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Committee on the Peaceful Uses of Outer Space

Information furnished in conformity with the Convention on Registration of Objects Launched into Outer Space

Note verbale dated 14 January 2021 from the Permanent Mission of Japan to the United Nations (Vienna) addressed to the Secretary-General

The Permanent Mission of Japan to the United Nations (Vienna), in accordance with article IV of the Convention on Registration of Objects Launched into Outer Space (General Assembly resolution 3235 (XXIX), annex), has the honour to transmit information concerning new and previously registered objects launched into outer space (see annex).[1]

[1] The data on the space object referenced in the annex were entered into the Register of Objects Launched into Outer Space on 28 January 2021.

Annex

Registration data on space objects launched by Japan*

Space Environment Data Acquisition Equipment – Attached Payload (SEDA-AP)

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1998-067PU
Name of space object	Space Environment Data Acquisition Equipment - Attached Payload (SEDA-AP)
State of registry	Japan
Other launching States	United States of America
Date and territory or location of launch	15 July 2009 UTC; Kennedy Space Center of the National Aeronautics and Space Administration, United States
Basic orbital parameters	
Nodal period	92.66 minutes
Inclination	51.64 degrees
Apogee	408.0 kilometres
Perigee	402.0 kilometres
General function of space object	This payload is a space environment monitoring facility at the International Space Station (ISS)

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	20 December 2018 at 2249 hours UTC
Space object owner or operator	Japan Aerospace Exploration Agency (JAXA)
Launch vehicle	STS-127 (Endeavour)
Other information	SEDA-AP was separated from ISS on 20 December 2018 at 2249 hours UTC SEDA-AP has no battery and is estimated to decay within 25 years

* The information was submitted using the form prepared pursuant to General Assembly resolution 62/101 and has been reformatted by the Secretariat.

Hagoromo

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1990-007B
Name of space object	Hagoromo
State of registry	Japan
Date and territory or location of launch	24 January 1990 UTC; Institute of Space and Astronautical Science (ISAS) Kagoshima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	Technology demonstration for lunar exploration

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	18 March 1990
Space object owner or operator	ISAS
Launch vehicle	M-3SII Launch Vehicle Flight No. 5
Celestial body space object is orbiting	Moon
Other information	Hagoromo was separated from the mother satellite Hiten (Muses-A) and moved to a lunar orbit on 18 March 1990 UTC

Minerva-II-1A

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	Minerva-II-1A
National designator/registration number	2014-076A-A
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-

Apogee	-
Perigee	-
General function of space object	Minerva-II-1A is a rover that landed on the surface of Ryugu and moved using a hopping mechanism

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	9 November 2019 at 1000 hours 1 second UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	Minerva-II-1A was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 21 September 2018 at 0405 hours UTC

Minerva-II-1B

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	Minerva-II-1B
National designator/registration number	2014-076A-B
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	Minerva-II-1B is a rover that landed on the surface of Ryugu and moved using a hopping mechanism

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	9 November 2019 at 1000 hours 1 second UTC

Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	Minerva-II-1B was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 21 September 2018 at 0405 hours UTC

OME-C1

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	OME-C1
National designator/registration number	2014-076A-C
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	OME-C1 is a cover that protects Minerva-II-1A and Minerva-II-1B

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	21 September 2018 at 0405 hours UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	OME-C1 was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 21 September 2018 at 0405 hours UTC

TM-B

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	TM-B
National designator/registration number	2014-076A-D
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	TM-B is a target marker for a touchdown operation by Hayabusa2

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	25 October 2018 UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	TM-B was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 25 October 2018 at 0237 hours UTC

DCAM3

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	DCAM3
National designator/registration number	2014-076A-G
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	

Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	DCAM3 is a deployable camera designed to observe operations by Hayabusa2

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	5 April 2019 at 0722 hours UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	DCAM3 was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 5 April 2019 at 0214 hours UTC

TM-A

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	TM-A
National designator/registration number	2014-076A-H
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	TM-A is a target marker for a touchdown operation by Hayabusa2

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	30 May 2019 UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	TM-A was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 30 May 2019 at 0218 hours UTC

TM-E

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	TM-E
National designator/registration number	2014-076A-K
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	TM-E is a target marker for a touchdown operation by Hayabusa2

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	23 September 2019 UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	TM-E was attached to asteroid explorer Hayabusa2, which was launched by an

	H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 16 September 2019 at 1617 hours UTC

TM-C

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	TM-C
National designator/registration number	2014-076A-L
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	TM-C is a target marker for a touchdown operation by Hayabusa2

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	23 September 2019 UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	TM-C was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 16 September 2019 at 1624 hours UTC

OME-C2

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	OME-C2
National designator/registration number	2014-076A-N
State of registry	Japan

Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	OME-C2 is a cover that protects Minerva-II-2

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	2 October 2019 at 1557 hours UTC
Space object owner or operator	JAXA
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	OME-C2 was attached to asteroid explorer Hayabusa2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 2 October 2019 at 1557 hours UTC

Minerva-II-2

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Name of space object	Minerva-II-2
National designator/registration number	2014-074A-M
State of registry	Japan
Date and territory or location of launch	3 December 2014 at 0422 hours 4 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	-
Inclination	-
Apogee	-
Perigee	-
General function of space object	Scientific observation and engineering demonstration of the hopping mechanism

	for an exploration robot on an asteroid surface with microgravity

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	9 November 2019 at 0303 hours 35 seconds UTC
Space object owner or operator	Tohoku University
Website	www.hayabusa2.jaxa.jp/en
Launch vehicle	H-IIA Launch Vehicle Flight No. 26
Celestial body	Ryugu
Other information	Minerva-II-2 was attached to asteroid explorer Hayabusa 2, which was launched by an H-IIA rocket on 3 December 2014. It was transferred to asteroid Ryugu by Hayabusa2 and deployed on 2 October 2019 at 1557 hours UTC

Super Low Altitude Test Satellite (SLATS) “Tsubame”

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2017-082B
Name of space object	Super Low Altitude Test Satellite (SLATS) “Tsubame”
State of registry	Japan
Registration document	ST/SG/SER.E/846
Date and territory or location of launch	23 December 2017 at 0126 hours 22 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	94.9 minutes
Inclination	98.3 degrees
Apogee	564.6 kilometres
Perigee	461.2 kilometres
General function of space object	SLATS will demonstrate orbit control operations at super low altitudes using ion engine technology developed by JAXA. Technical data related to the atmosphere acquired by SLATS will also be used for the design of future satellites Furthermore, SLATS will photograph the Earth and its technology will be evaluated for future Earth observation satellites

Date of decay/re-entry/deorbit	1 October 2019 at 1013 hours 0 seconds UTC

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	JAXA
Launch vehicle	H-IIA Launch Vehicle Flight No. 37
Other information	Launching organizations are Mitsubishi Heavy Industries, Ltd. and JAXA Basic orbital parameters described are as at 25 January 2018 In the future, orbit control operations will be used to lower the altitude sequentially

H-II Transfer Vehicle “Kounotori 8” (HTV8)

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2019-062A
Name of space object	H-II Transfer Vehicle “Kounotori 8” (HTV8)
State of registry	Japan
Date and territory or location of launch	24 September 2019 at 1605 hours 05 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	92.9 minutes
Inclination	51.6 degrees
Apogee	358.5 kilometres
Perigee	346.0 kilometres
General function of space object	HTV8 is an uncrewed resupply vehicle designed to transport various pieces of cargo including research materials, replacement equipment and daily commodities to ISS
Date of decay/reentry/deorbit	3 November 2019 UTC

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	JAXA
Launch vehicle	H-IIB Launch Vehicle Flight No. 8
Other information	Launching organizations are Mitsubishi Heavy Industries, Ltd. and JAXA Basic orbital parameters described are as at 28 September 2019

	After delivering its cargo to ISS, HTV8 will be unberthed from ISS and will make a controlled re-entry into the atmosphere

AES Satellite “SOCRATES”

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2014-029C
Name of space object	Advanced Engineering Services (AES) Satellite “SOCRATES”
National designator/registration number	2014-029C
State of registry	Japan
Registration document	ST/SG/SER.E/735
Date and territory or location of launch	24 May 2014 at 0305 hours 14 seconds UTC; JAXA Tanegashima Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	97.2 minutes
Inclination	97.9 degrees
Apogee	628.9 kilometres
Perigee	618.4 kilometres
General function of space object	Demonstration of the small satellite standard bus and provision of an environment to demonstrate advanced missions and element technologies in orbit

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	4 June 2019 at 0307 hours 27 seconds UTC;
Physical conditions when space object is moved to a disposal orbit	After the mission was completed, the satellite stopped functioning by telecommand. The satellite was made safe by separating the battery from the load
Space object owner or operator	AES
Launch vehicle	H-IIA Launch Vehicle Flight No. 24 (H-IIA 24F)
Other information	Launching organizations are Mitsubishi Heavy Industries, Ltd. and JAXA Basic orbital parameters described are as at 30 June 2014

RSP-00

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1998-067PP
Name of space object	RSP-00
State of registry	Japan
Date and territory or location of launch	6 October 2018 at 1700 hours 0 seconds UTC; ISS
Basic orbital parameters	
Nodal period	91 minutes
Inclination	51.6 degrees
Apogee	401.8 kilometres
Perigee	393.7 kilometres
General function of space object	Technology demonstration of a transmitter that realizes higher speed transmissions than a conventional transmitter by sending photos of the Earth taken by RSP-00 itself. A conventional transmitter is also installed and sends photos

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations	
Date when space object is no longer functional	6 October 2018 UTC
Space object owner or operator	Ryman Sat Project Japan
Other information	Launched by H-IIB F7 on 22 September 2018 UTC. RSP-00 was carried on HTV-7 and transported to ISS Date of launch is the date of deployment from ISS and territory or location of launch is the location of deployment

SPATIUM-I

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1998-067PN
Name of space object	SPATIUM-I
State of registry	Japan
Date and territory or location of launch	6 October 2018 UTC; ISS

Basic orbital parameters	
Nodal period	92.4 minutes
Inclination	51.6 degrees
Apogee	393 kilometres
Perigee	388 kilometres
General function of space object	Demonstration of an on-board chip-scale atomic clock (CSAC) and spread spectrum transmission using CSAC as the clock source Time-synchronization of multiple ground stations Reading of the carrier wave phases of a single satellite

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	Kyushu Institute of Technology, Japan
Website	www.facebook.com/Space-Precision-Atomic-clock-TIming-Utility-Mission-293774767872332/?modal=admin_todo_tour
Other information	Launched by an H-IIB rocket on 22 September 2018 and carried to ISS by the HTV-7 spacecraft Date of launch is the date of deployment from ISS and territory or location of launch is the location of deployment

Toki

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1998-067MU
Name of space object	Toki
State of registry	Japan
Registration document	ST/SG/SER.E/862
Date and territory or location of launch	7 July 2017 UTC; ISS
Basic orbital parameters	
Nodal period	91.7 minutes
Inclination	51.6 degrees
Apogee	359 kilometres
Perigee	357 kilometres
General function of space object	Earth observation, outreach by sound signal transmission and single event detection
Date of decay/re-entry/deorbit	3 May 2019 UTC

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	Kyushu Institute of Technology, Japan
Website	birds1.birds-project.com
Other information	Launched by Falcon 9 rocket on 4 June 2017 and carried to ISS by the SpaceX Dragon CRS-11 spacecraft Date of launch is the date of deployment from ISS and territory or location of launch is the location of deployment

Uguisu

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1998-067QG
Name of space object	Uguisu
State of registry	Japan
Date and territory or location of launch	17 June 2019 UTC; ISS
Basic orbital parameters	
Nodal period	91.1 minutes
Inclination	51.6 degrees
Apogee	416 kilometres
Perigee	415 kilometres
General function of space object	Short message transmission by continuous beacon, Earth observation by camera module, measurement of the geomagnetic field, posture stabilization, on-orbit operation demonstration of LoRa module and on-orbit operation demonstration of a complex programmable logic device

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	Kyushu Institute of Technology, Japan
Website	birds3.birds-project.com
Other information	Launched by Antares rocket on 17 April 2019 and carried to ISS by the Cygnus NG-11 spacecraft Date of launch is the date of deployment from ISS and territory or location of launch is the location of deployment

NEXUS

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2019-003F
Name of space object	NEXUS
State of registry	Japan
Date and territory or location of launch	18 January 2019 at 0050 hours 20 seconds UTC; JAXA Uchinoura Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	93.5 minutes
Inclination	97.3 degrees
Apogee	508.5 kilometres
Perigee	488.5 kilometres
General function of space object	NEXUS is a 10 cm cubic amateur satellite. The aim of the mission is to give a demonstration using transmitters and a linear transponder in the space environment

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	Nihon University
Launch vehicle	Epsilon Launch Vehicle Flight No. 4
Other information	Launched by JAXA

STARS-Me

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	1998-067PQ
Name of space object	STARS-Me
State of registry	Japan
Date and territory or location of launch	6 October 2018 at 0800 hours 0 seconds UTC; ISS
Basic orbital parameters	
Nodal period	92.5 minutes
Inclination	51.639 degrees
Apogee	393 kilometres
Perigee	401 kilometres
General function of space object	STARS-Me consists of two 1U CubeSats that have independent basic functionality,

	each satellite communicating independently with the ground station. The two CubeSats are connected by a tether. STARS-Me is first secured together and put into orbit followed by deployment of the tether resulting in the separation of the two satellites. Thereafter, a “climber” will traverse the deployed tether. The climber, using Bluetooth, will transmit data to a ground station via a STARS-Me CubeSat

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	Shizuoka University
Website	stars.eng.shizuoka.ac.jp/english.html
Other information	Launched by an H-IIB rocket on 22 September 2018 and carried to ISS by the HTV-7 spacecraft Date of launch is the date of deployment from ISS and territory or location of launch is the location of deployment

N-SAT-110

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2000-060A
Name of space object	N-SAT-110
State of registry	Japan
Other launching States	France
Registration document	ST/SG/SER.E/407
Date and territory or location of launch	6 October 2000 at 2300 hours UTC; Guiana Space Centre, Kourou, French Guiana
Basic orbital parameters	
Nodal period	1,436 minutes
Inclination	0.029 degrees
Apogee	35,797 kilometres
Perigee	35,779 kilometres
General function of space object	Domestic communications and domestic broadcasting
Date of decay/re-entry/deorbit	10 January 2019 at 1055 hours UTC

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Change of status in operations

Date when space object is no longer functional	17 January 2019 at 0137 hours UTC
Date when space object is moved to a disposal orbit	10 January 2019 at 1055 hours UTC
Physical conditions when space object is moved to a disposal orbit	Satellite achieved a disposal altitude of 290 km above the geostationary orbit and all satellite systems were shut down Fuel deplete operation and battery charge terminate operation were executed without incident
Geostationary position	110 degrees East
Space object owner or operator	SKY Perfect JSAT Corporation
Launch vehicle	Ariane 42L
Other information	Launch organization is Arianespace

Epsilon Launch Vehicle Flight No. 4 rocket body (third stage)

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2019-003H
Name of space object	Epsilon Launch Vehicle Flight No. 4 rocket body (third stage)
State of registry	Japan
Date and territory or location of launch	18 January 2019 at 0050 hours 20 seconds UTC; JAXA Uchinoura Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	94 minutes
Inclination	97.18 degrees
Apogee	721 kilometres
Perigee	220 kilometres
General function of space object	This space object is part of the spent rocket body of Epsilon Launch Vehicle Flight No. 4. The launch vehicle consists of a third stage and a post-boost stage; this object is the third stage
Date of decay/re-entry/deorbit	7 August 2019 UTC

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	JAXA
Launch vehicle	Epsilon Launch Vehicle Flight No. 4
Other information	Launching organization is JAXA. The rocket body (third stage) has no stored energy at the end of combustion (solid

	rocket motor) and is estimated to decay within 25 years
	Basic orbital parameters were as at separation of the third stage from Epsilon Launch Vehicle Flight No. 4

Epsilon Launch Vehicle Flight No. 4 rocket body (post-boost stage)

Information provided in conformity with the Convention on Registration of Objects Launched into Outer Space

Committee on Space Research international designator	2019-003E
Name of space object	Epsilon Launch Vehicle Flight No. 4 rocket body (post-boost stage)
State of registry	Japan
Date and territory or location of launch	18 January 2019 at 0050 hours 20 seconds UTC; JAXA Uchinoura Space Center, Kagoshima, Japan
Basic orbital parameters	
Nodal period	94 minutes
Inclination	97.33 degrees
Apogee	500 kilometres
Perigee	482 kilometres
General function of space object	This space object is part of the spent rocket body of Epsilon Launch Vehicle Flight No. 4. The launch vehicle consists of a third stage and a post-boost stage; this object is the post-boost stage

Additional voluntary information for use in the Register of Objects Launched into Outer Space

Space object owner or operator	JAXA
Launch vehicle	Epsilon Launch Vehicle Flight No. 4
Other information	Launching organization is JAXA. The post-boost stage is designed to release stored pressure upon passivation and is estimated to decay within 25 years
	Basic orbital parameters were acquired just prior to loss of communication with the post-boost stage